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## Abstract

When a mission for multistage rocket is fixed, the problem of optimum staging for rocket design comes in. In the proposed study we have discussed the treatment of optimization problem under two subsidiary conditions of restriction, namely for a constant initial gross mass of the rocket, to find optimum stage-mass distribution to achieve minimum propellant mass for a given required payload and burnout velocity. The usual assumption of drag free and gravitation less

environment is taken.

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Computer Science

## Index Terms

Multistage Rockets

## Keywords

Propellant Factors Exhaust Velocities And Optimum Stage Mass Distribution Of Rocket

Lagrange's Multipliers